Claims

1. Combustion chamber (4) for a gas turbine (1), the combustion space (24) of which is bounded by an annular combustion chamber inner wall (28) and an annular combustion chamber outer wall (26) which are provided on their inside with a lining formed from a plurality of heat shield elements (38), wherein the or each heat shield element (28) forms together with the combustion chamber wall (25) an inner space (40) to which a cooling medium (K) can be applied and in which there is disposed a cooling medium distributor (42) and wherein the combustion chamber inner wall (28) is formed from a plurality of wall elements (30) abutting each other at a horizontal parting joint (31), said wall elements (30) being connected to each other in the area of the parting joint (31) by means of a plurality of screw connections (32) oriented at an angle to the inner wall surface.

2. Combustion chamber (4) according to claim 1, wherein a feather key (35) is assigned to the or each screw connection (32).

3. Combustion chamber (4) according to claim 1, wherein a cooling medium supply line (44) is connected to a plurality of cooling medium exit openings (46) via a cooling medium distributor (42).

4. Combustion chamber (4) according to claims 1 to 3, wherein the cooling medium exit openings (46) are dimensioned such that the sum total of the cross-sectional areas of all the cooling medium exit openings (46) of a cooling medium distributor (42) is less than the cross-sectional area of the assigned cooling medium supply line (44).

5. Combustion chamber (4) according to one of claims 1 to 4, wherein the or each inner space (40) is connected to a cooling medium discharge system via a plurality of holes.

6. Combustion chamber (4) according to claim 1, wherein the heat shield elements (38) are fixed to the combustion chamber inner wall (28) or to the combustion chamber outer wall (26) via a tongue and groove system.

7. Gas turbine (1) with a combustion chamber (4) according to one of claims 1 to 5.